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***In the Claims***

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This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims

1. (Currently amended) A process for removing a thermal barrier ceramic coating from a metallic substrate surface of a component comprising: directing an air jet at the thermal barrier coating on the substrate surface of the component, the jet containing a non-abrasive particulate media and ~~being emitting~~ the media from a nozzle of the jet at a low pressure wherein said low pressure is insufficient ~~to~~ for the media to damage the substrate surface but said low pressure is sufficient ~~to~~ for the media to remove the thermal barrier ceramic coating.
2. (Currently amended) The [P]process of claim 1 wherein the pressure of the air jet is from about 20 to 100 PSIG.
3. (Currently amended) The [P]process of claim 2 wherein the media has a substantially spherical shape.
4. (Currently amended) The [P]process of claim 3 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
5. (Currently amended) The [P]process of claim 4 wherein the media is glass beads.
6. (Currently amended) The [P]process of claim 1 wherein the component is a turbine engine component.
7. (Currently amended) The [P]process of claim 6 wherein the turbine engine component is a combustion chamber.
8. (Currently amended) A process for removing a thermal barrier ceramic coating from a cooling hole of a metallic turbine engine component comprising: directing an air jet at the cooling hole of the component, the jet containing non-abrasive particulate media and ~~being emitting~~ the media from a nozzle of the jet at a low pressure wherein said low pressure is insufficient ~~to~~ for the media to damage a metallic surface of the cooling hole but said low pressure is sufficient ~~to~~ for the media to remove the thermal barrier ceramic coating.
9. (Currently amended) The [P]process of claim 8 wherein the pressure of the air jet is from about

20 to 100 PSIG

10. (Currently amended) The [P]process of claim 9 wherein the media has a substantially spherical shape.
11. (Currently amended) The [P]process of claim 10 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches
12. (Currently amended) The [P]process of claim 11 wherein the media is glass beads.
13. (Currently amended) The [P]process of claim 12 wherein the turbine engine component is a combustion chamber.
14. (Currently amended) The [P]process of claim 8 wherein the air jet is directed at the cooling hole toward a surface of the component opposing the surface having the thermal barrier coating.
15. (Currently amended) The [P]process of claim 9 wherein the air jet is directed at the cooling hole at substantially the same angle as the cooling hole.
16. (Currently amended) The [P]process of claim 8 wherein the air jet rounds the metallic edges of the cooling hole.
17. (Currently amended) The [P]process of claim 8 wherein the cooling holes are drilled into the turbine component using a laser drilling process.
18. (Currently amended) A process for forming cooling holes on a thermal barrier coated turbine engine component comprising: drilling cooling holes into the component; coating a surface of the component containing the cooling holes with a thermal barrier ceramic coating; and directing an air jet at the cooling hole of the component, the jet containing non-abrasive particulate media and being emitted the media from a nozzle of the jet at a low pressure wherein said low pressure is insufficient to for the media to damage a metallic surface of the cooling hole but said low pressure is sufficient to for the media to remove the thermal barrier coating.
19. (Currently amended) The [P]process of claim 18 wherein the pressure of the air jet is from about 20 to 100 PSIG
20. (Currently amended) The [P]process of claim 19 wherein the media has a substantially spherical shape.

21. (Currently amended) The [P]process of claim 20 wherein the spherical media particles have a diameter of from about 0.002 to 0.010 inches.
22. (Currently amended) The [P]process of claim 21 wherein the media is glass beads.
23. (Currently amended) The [P]process of claim 22 wherein the turbine engine component is a combustion chamber.
24. (Currently amended) The [P]process of claim 16 wherein the air jet is directed at the cooling hole toward a surface of the component opposing the surface having the thermal barrier coating.
25. (Currently amended) The [P]process of claim 18 wherein the air jet is directed at the cooling hole at substantially the same angle as the cooling hole.
26. (Currently amended) The [P]process of claim 18 wherein the air jet rounds the metallic edges of the cooling hole
27. (Currently amended) The [P]process of claim 18 wherein the cooling holes are drilled through the turbine component using a laser drilling process.